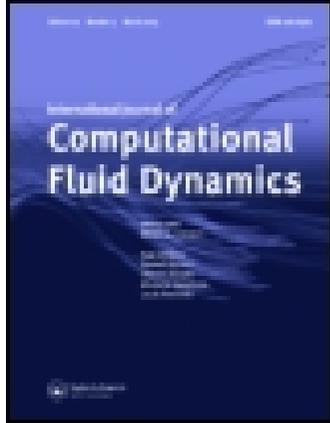


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## Evaluation of linear, inviscid, viscous, and reduced-order modelling aeroelastic solutions of the AGARD 445.6 wing using root locus analysis

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Reduced-order modelling (ROM) methods are applied to the Computational Fluid Dynamics (CFD)-based aeroelastic analysis of the AGARD 445.6 wing in order to gain insight regarding well-known discrepancies between the aeroelastic analyses and the experimental results. The results presented include aeroelastic solutions using the inviscid Computational Aeroelasticity Programme–Transonic Small Disturbance (CAP-TSD) code and the FUN3D code (Euler and Navier–Stokes). Full CFD aeroelastic solutions and ROM aeroelastic solutions, computed at several Mach numbers, are presented in the form of root locus plots in order to better reveal the aeroelastic root migrations with increasing dynamic pressure. Important conclusions are drawn from these results including the ability of the linear CAP-TSD code to accurately predict the entire experimental flutter boundary (repeat of analyses performed in the 1980s), that the Euler solutions at supersonic conditions indicate that the third mode is always unstable, and that the FUN3D Navier–Stokes solutions stabilize the unstable third mode seen in the Euler solutions.

**Keywords:** ROM; aeroelasticity; flutter; CFD; AGARD 445.6

### 1. Introduction

Classical linear aeroelastic analyses typically produce velocity–damping–frequency (V-g-f) plots and/or root locus plots. The use of these plots has enabled the aeroelastician to view the nature of the flutter mechanism(s) in addition to identifying the condition(s) at which flutter occurs. The rapid creation of these plots was facilitated by the use of linear unsteady aerodynamics and linear aeroelastic equations of motion (Adams and Hoadley 1993).

During the last few years, higher order CFD-based methods have become an important method for the computation of nonlinear unsteady aerodynamics for use in aeroelastic analyses. The use of these higher order methods provides valuable insight regarding complex flow physics at conditions where linear methods are not theoretically valid. However, the increased computational cost associated with the computation of unsteady aerodynamics and aeroelastic responses using higher order methods has resulted in a subtle change in the manner in which the aeroelastician evaluates and interprets these analyses. First, the increased computational cost of these analyses has tended to dictate a ‘snapshot’ approach to aeroelastic analyses whereby the aeroelastic response at a handful of dynamic pressures is all that is computed. This ‘snapshot’ approach is used to identify the flutter dynamic pressure but the actual flutter mechanism is not easily discernible. Second, due to the complexity of the computational methods, methods that could

rapidly generate V-g-f plots and/or root locus plots were not available. However, with the development of reduced-order modelling (ROM) methods (Silva 2008; Silva, Vatsa, and Biedron 2009, 2010), the rapid generation of root locus plots using CFD-based unsteady aerodynamics is now available to aeroelasticians.

The goal behind the development of a ROM for the rapid computation of unsteady aerodynamic and aeroelastic responses is aimed at addressing two challenges. The first challenge is the computational cost associated with full CFD aeroelastic simulations, which increases with the fidelity of the nonlinear aerodynamic equations to be solved as well as the complexity of the configuration. Computational cost, however, may be reduced via the implementation of parallel processing techniques, advanced algorithms, and improved computer hardware processing speeds.

The second, more serious, challenge is that the information generated by these simulations cannot be used effectively within a preliminary design environment. For this reason, parametric variations and design studies can only be performed by trial-and-error. As a result, the integration of computational aeroelastic simulations into preliminary design activities involving disciplines such as aeroelasticity, aeroservoelasticity (ASE), and optimization continues to be a costly and impractical venture.

Development of a ROM entails the development of a simplified mathematical model that captures the dominant

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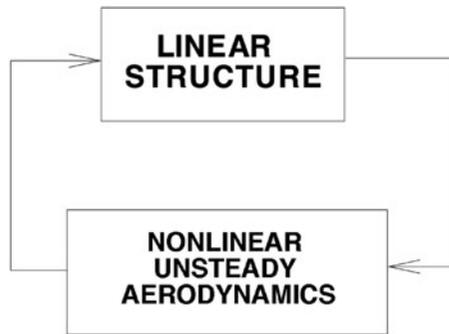


Figure 1. Coupling of linear structural model and nonlinear unsteady aerodynamics within an aeroelastic CFD code such as FUN3D.

dynamics of the original system. This alternative mathematical representation of the original system is, by design, in a mathematical form suitable for use in a multidisciplinary, preliminary design environment. As a result, use of the ROM in other disciplines is possible, thereby addressing the second challenge. The simplicity of the ROM yields significant improvements in computational efficiency as compared to the original system, thereby, also addressing the first challenge.

A CFD-based model of an aeroelastic system (such as the FUN3D code (Anderson and Bonhaus 1994; Biedron and Thomas 2009a)) consists of the coupling of a nonlinear unsteady aerodynamic system (flow solver) with a linear structural system as depicted in Figure 1. Standard CFD-based aeroelastic analyses are performed via iterations between the nonlinear unsteady aerodynamic system and the linear structural system. Throughout this paper, these standard CFD-based solutions will be referred to as full CFD solutions’.

The present study also involves the generation of linearized unsteady aerodynamic ROMs (in state-space form), using the general procedure depicted in Figure 2. In this situation, the linear structural system within the CFD code is bypassed so that only the nonlinear unsteady aerodynamic system is excited. Specific modal inputs are applied to the nonlinear unsteady aerodynamic system and the gen-

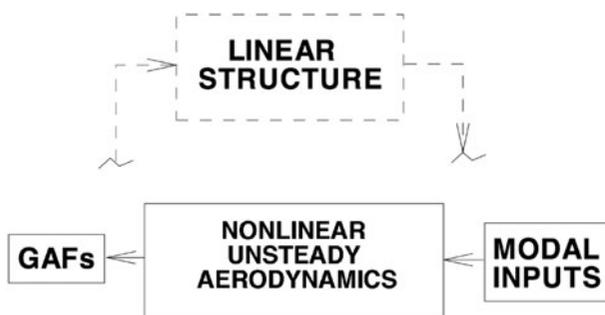


Figure 2. Generation of generalized aerodynamic forces (GAFs) used for system identification process.

eralized aerodynamic force (GAF) outputs from this solution, along with the inputs, are used in a system identification process to create the linearized unsteady aerodynamic ROM. This unsteady aerodynamic ROM is then coupled to a state-space model of the structure in order to create the aeroelastic simulation ROM. The aeroelastic simulation ROM is then used for aeroelastic analyses. For the discussions that follow, the term ROM will refer to the unsteady aerodynamic state-space model. When the unsteady aerodynamic state-space model (ROM) is connected to a state-space model of the structure, this system is often also referred to as a ROM. However, to avoid confusion, the aeroelastic system consisting of an unsteady aerodynamic ROM (state-space form) coupled with a linear modal model of the structure (also state-space form) will be referred to as the aeroelastic simulation ROM.

The development of CFD-based ROMs continues to be an area of active research at several national and international government, industry, and academic institutions (Silva 2005; Silva and Bartels 2004; Beran and Silva 2001; Kim et al. 2005; Raveh 2004). Development of ROMs based on the Volterra theory is one of several ROM methods that has received attention over the last few years (Silva et al. 2001; Silva 1999, 1997, 1993; Raveh, Levy, and Karpel 2000; Balajewicz, Nitzche, and Feszty 2009; Omran and Newman 2009; Milanese and Marzocca 2009). Although the primary focus of this paper is the development and application of unsteady aerodynamic ROMs for subsequent use in aeroelastic analyses, the development of ROMs for the rapid computation of nonlinear stability and control derivatives using CFD codes (Jirasek and Cummings 2009; Ghoreyshi et al. 2012) is an active area of research as well.

Silva and Bartels (2004) introduced the development of linearized, unsteady aerodynamic state-space models for prediction of flutter and aeroelastic response using the parallelized, aeroelastic capability of the CFL3Dv6 code. The results presented provided an important validation of the various phases of the ROM development process. The eigensystem realization algorithm (ERA) (Juang and Pappa 1985), which transforms an impulse response (one form of a ROM) into state-space form (another form of a ROM), was applied to the development of aerodynamic state-space models. The ERA is a part of the SOCIT (System/Observer/Controller Identification Toolbox) (Juang 1994). Flutter results for the AGARD 445.6 aeroelastic wing using the CFL3Dv6 code were presented, including computational costs (Silva and Bartels 2004). Unsteady aerodynamic state-space models were generated and coupled with a structural model within a MATLAB/SIMULINK (MathWorks XXXX) environment for rapid calculation of aeroelastic responses including the prediction of flutter. Aeroelastic responses computed directly using the aeroelastic simulation ROM showed excellent comparison with the aeroelastic responses computed using the CFL3Dv6 code (Krist, Biedron, and Rumsey 1997).

Previously (Silva and Bartels 2004), the aerodynamic impulse responses (unit pulses) that were used to generate the unsteady aerodynamic state-space model were computed via the excitation of one mode at a time. For a four-mode system such as the AGARD 445.6 wing, these computations are not very expensive. However, for more realistic cases where the number of modes can be an order of magnitude or more larger, the one-mode-at-a-time method becomes prohibitively expensive. Towards the solution of this problem, new methods have been developed. Kim et al. (2005) have proposed methods that enable the simultaneous application of structural modes as CFD input, greatly reducing the cost of identifying the aerodynamic impulse responses from the CFD code. Kim's method consists of using simultaneous staggered step inputs, one per mode, and then recovering the individual responses from this simultaneous excitation. Silva (2008) has developed a method that enables the simultaneous excitation of the structural modes using orthogonal functions. Both of these methods require only a single CFD solution and the methods are independent of the number of structural modes. Silva (2007) has also developed a method for generating static aeroelastic solutions and matched-point aeroelastic solutions using a ROM. The methods developed by Silva (2008, 2007) have already been implemented in the FUN3D CFD code. In addition, methods for generating root locus plots of the combined structural state-space model and unsteady aerodynamic state-space model were developed by Silva, Vatsa, and Biedron (2009). These ROM-based root locus methods were applied to fixed-wing configurations and subsequently to launch vehicle configurations (Silva, Vatsa, and Biedron 2010). The present paper will focus on the application of these ROM and root locus methods in order to visualize the aeroelastic behaviour of the AGARD 445.6 wing as a function of Mach number, dynamic pressure, and fluid dynamic equation (Computational Aeroelasticity Programme–Transonic Small Disturbance (CAP-TSD), inviscid FUN3D, and viscous FUN3D).

The paper begins with a description of the AGARD 445.6 wing and a comparison of experimental and computational flutter results obtained to date by various researchers (Chwalowski et al. 2011). Computational methods and related models are introduced including the CAP-TSD code, the FUN3D code (inviscid and viscous grids), and the FUN3D ROM creation process. The results to be presented are grouped into two categories. The first category consists of the full CFD solutions' based on the standard iterative approach briefly described above. These results will include full CAP-TSD solutions and full FUN3D solutions (inviscid and viscous) for several Mach numbers and dynamic pressures. The aeroelastic transients computed via the full CFD solutions are analysed for their damping and frequency content in order to generate aeroelastic root locus plots. The second category of results consists of the 'FUN3D ROM solutions'. At present, the ROM method

has not been implemented in the CAP-TSD code; therefore, all ROM solutions will be FUN3D ROM solutions. The FUN3D ROM solutions will be presented in the form of root locus plots at several Mach numbers. Finally, some concluding remarks will be provided.

## 2. AGARD 445.6 wing

The AGARD 445.6 wing was tested in the NASA Langley Transonic Dynamics Tunnel (TDT) in 1961 (Yates et al. 1963). Flutter data from this test have been publicly available for over 20 years and have been widely used for preliminary computational aeroelastic benchmarking. The AGARD wing planform was sidewall-mounted and had a quarter-chord sweep angle of  $45^\circ$ , an aspect ratio of 1.65, a taper ratio of 0.66, a wing semi-span of 2.5 feet, a wing root chord of 1.833 feet, and a symmetric airfoil. The wing was flutter tested in both air and R-12 heavy gas test mediums at Mach numbers from 0.34 to 1.14 at  $0^\circ$  angle of attack. Unfortunately, this data-set lacks unsteady surface pressure measurements necessary for more extensive code validation.

A broad range of FUN3D computations (Chwalowski et al. 2011) for the AGARD 445.6 wing was performed across the entire Mach number range of the experimental data, assuming both inviscid and viscous flows with air as the working fluid. The first four structural modes were used in the aeroelastic analysis and are presented in Figure 3.

Figures 4 and 5 present comparisons among the experimental flutter speed index and frequency ratio values, respectively, with those obtained using the FUN3D code, and those published in the literature (Lee-Rausch and Batina 1993, 1995; Gupta 1996; Pahlavanloo 2007). In the figure legends, the inviscid results are the Euler (E) results, and the viscous results are the Navier–Stokes (NS) results. The SA represents the Spalart–Allmaras turbulence model. In general, in the subsonic flow regime, the computational data match the experimental data well, while a broad range in the computational data is observed in the high subsonic and supersonic flow regimes. Chwalowski et al. (2011) emphasize the importance of applying the viscous flow assumption (using FUN3D) at the high subsonic conditions as evidenced by the improved result over the FUN3D inviscid subsonic solutions. In addition, Chwalowski et al. (2011) carried out a grid refinement study that improved the correlation between the viscous FUN3D solution and the experiment at the supersonic Mach numbers. However, an important point to be made is that the FUN3D (inviscid and viscous) flutter solutions at the supersonic Mach numbers are for a flutter mechanism which comprised the coalescence of the first and second modes. Based on the frequency ratio of the results presented from other references, it appears that those results correspond to the same mechanism. This assumption needs to be confirmed with each individual researcher

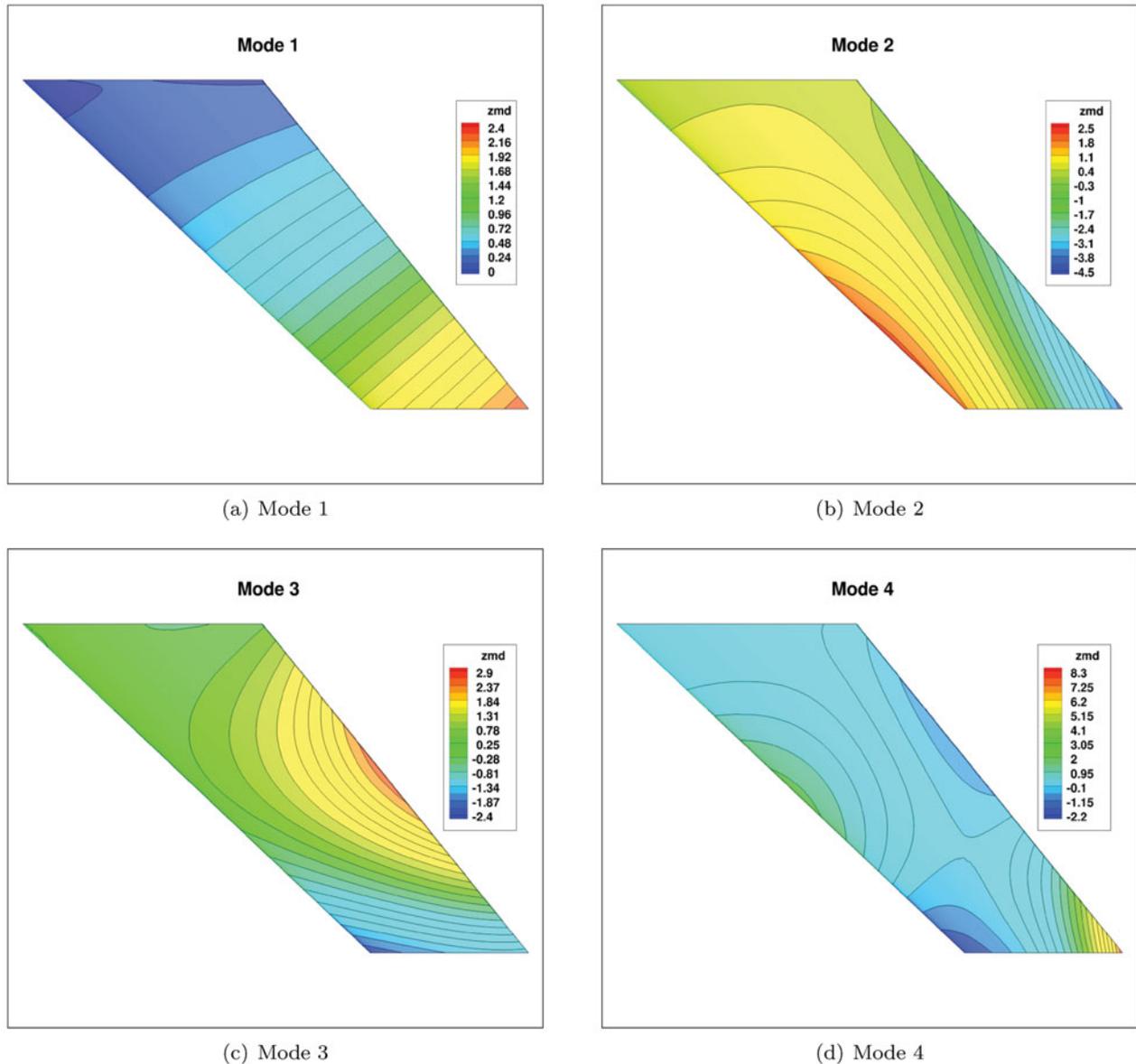


Figure 3. The first four modes of the AGARD 445.6 wing where ‘zmd’ is the modal deflection in the  $z$ -direction.

that has provided the results for this wing included in these figures.

In order to simplify the discussion regarding comparisons of experimental and various full CFD and ROM solutions, our results will be presented in terms of flutter dynamic pressure (psf) and flutter frequency (Hz) for a subset of FUN3D results only. Figure 6 presents flutter boundaries in terms of flutter dynamic pressure in psf for experiment, FUN3D/Euler (FUN3D-E), FUN3D/Navier–Stokes/Spallart–Allmaras turbulence model for the baseline grid (FUN3D NS SA Baseline Grid), and FUN3D/Navier–Stokes/Spallart–Allmaras turbulence model for the fine grid (FUN3D NS SA Fine Grid). The comparison between the baseline grid and the fine grid is the result of the grid

refinement study previously mentioned. Figure 7 presents the corresponding flutter frequencies (Hz) as a function of Mach number. All subsequent FUN3D results in this paper will include Euler solutions and NS solutions (with the SA turbulence model) for the baseline grid.

As will be discussed in the subsequent sections of this paper, the discrepancy between the computational and experimental results at the supersonic Mach numbers raises interesting questions. The first question concerns results obtained by Yates (1988) using modified strip analysis as well as results obtained by Bennett, Batina, and Cunningham (1989) in the late 1980s using the CAP-TSD code (linear and nonlinear solutions). Both references indicate that the entire flutter boundary is well predicted using

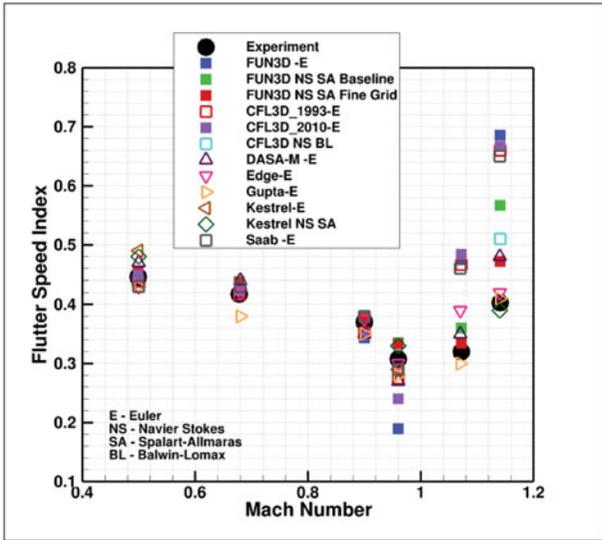


Figure 4. Experimental and computational flutter speed index versus Mach number for the AGARD 445.6 wing.

linear methods. These results are consistent with the fact that the AGARD 445.6 wing has a thin airfoil and the wing does not reach transonic conditions until about  $M = 0.98$ . At the supersonic Mach numbers of interest ( $M = 1.07$  and  $M = 1.14$ ), the flow is entirely supersonic and should therefore be well predicted with the use of linear unsteady aerodynamic methods. Given these facts, the first question is as follows: why are the Euler and NS solutions so different from the linear solutions at conditions where there is, essentially, no nonlinearity in the flow? Furthermore, the flutter results computed using the linear equations within the CAP-TSD code serve to correct statements often found in other references that classify the flutter boundary of the

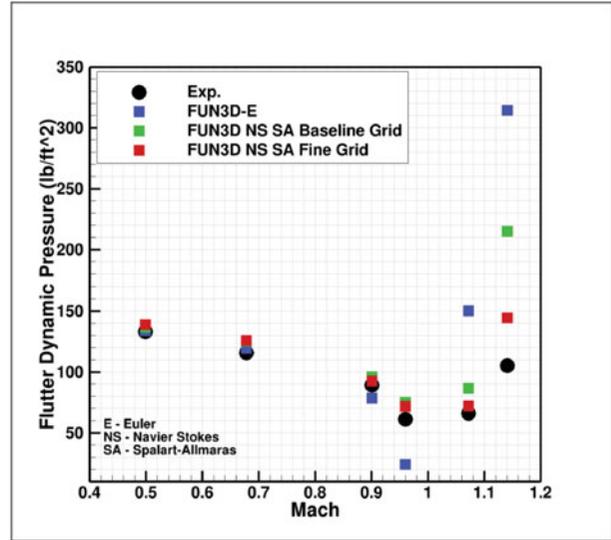


Figure 6. Experimental and computational flutter dynamic pressure (psf) versus Mach number for the AGARD 445.6 wing.

AGARD 445.6 wing as a nonlinear transonic flutter dip. Clearly, if linear CAP-TSD can accurately predict the flutter boundary at high subsonic Mach numbers and if transonic flow is not present at Mach numbers below  $M = 0.98$ , the flutter boundary up to that Mach number cannot be referred to as a transonic flutter dip. The dip in the flutter boundary that is observed is due, primarily, to compressibility and is not the result of a nonlinear transonic effect.

The second question raised by the discrepancy in the computational and experimental flutter results at the supersonic Mach numbers is as follows: what is the actual flutter mechanism at supersonic conditions? The experimental value of flutter frequency at  $M = 1.141$  is

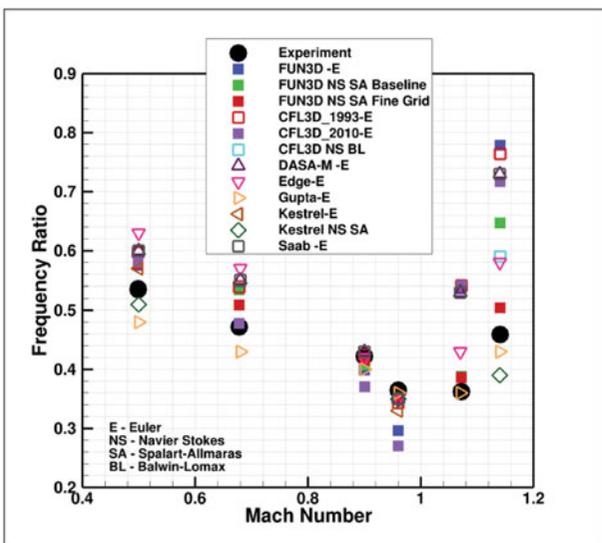


Figure 5. Experimental and computational frequency ratio versus Mach number for the AGARD 445.6 wing.

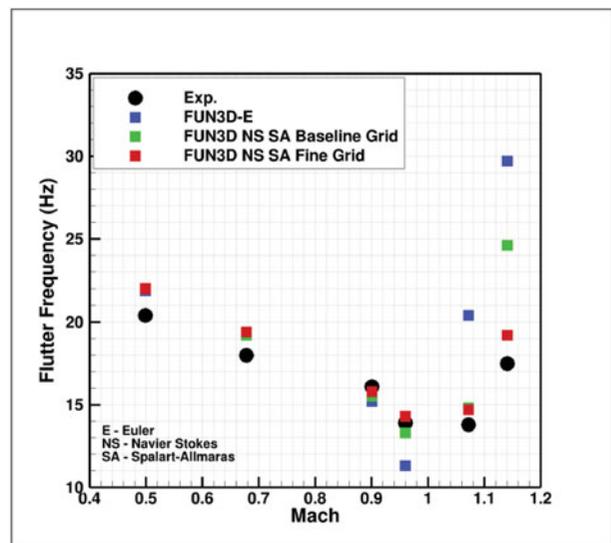


Figure 7. Experimental and computational flutter frequency (Hz) versus Mach number for the AGARD 445.6 wing.

about 17.5 Hz, suggesting that the measured flutter mechanism involved the first and second modes. Based on this indication, it appears that computational results presented to date have focused on this particular flutter mechanism. However, it will be shown later in the paper that the third mode is always unstable for the inviscid (Euler) solutions. It is not clear if this situation is true for all other inviscid solutions performed by other researchers but it is certainly true for the inviscid solutions computed using FUN3D. The authors have confirmed that this is the case as well for the inviscid solution from the CFL3Dv6 code (a NASA Langley-developed structured CFD aeroelastic code), although those results are not presented here. The use of FUN3D ROMs at this condition served to highlight this issue with the obvious clarity of a root locus plot to be presented later in the paper. Although this discussion may raise more questions than it answers, the authors feel this is an important discussion to be raised if we are to seriously address the validation of our computational aeroelastic tools.

### 3. Computational methods

#### 3.1. CAP-TSD code

The CAP-TSD code is a finite difference programme that solves the general-frequency modified TSD potential equation. The TSD potential equation is solved within CAP-TSD by a time-accurate approximate factorization (AF) algorithm developed by Batina (1987). The CAP-TSD programme can be used for the analysis of configurations with combinations of lifting surfaces and bodies including canard, wing, tail, control surfaces, tip launchers, pylons, fuselage, stores, and nacelles. The CAP-TSD code was applied to several configurations (Cunningham, Batina, and Bennett 1988; Bennett, Batina, and Cunningham 1989; Silva and Bennett 1995) in the late 1980s and the early 1990s.

In the present effort, linear CAP-TSD solutions were generated, repeating the work performed in the late 1980's. A new contribution to that original effort was the inclusion of root locus plots generated from the CAP-TSD aeroelastic transient responses. The procedure for generating the aeroelastic transients at various Mach numbers and dynamic pressures using CAP-TSD is the same as the procedure for FUN3D described in a subsequent section.

#### 3.2. FUN3D code, grids, and analysis procedure

The following subsections describe the parallelized, aeroelastic version of the unstructured mesh solver FUN3D code, the inviscid and viscous grids used, and a brief review of what will be referred to as the FUN3D full solution (in contrast to the FUN3D ROM solution).

##### 3.2.1. FUN3D code

The unstructured mesh Euler (inviscid)/NS (viscous) solver used for this study is FUN3D (Anderson and Bonhaus

1994). Within the code, the unsteady NS equations are discretized over the median dual volume surrounding each mesh point, balancing the time rate of change of the averaged conserved variables in each dual volume with the flux of mass, momentum, and energy through the instantaneous surface of the control volume. Additional details regarding the aeroelastic capability within the FUN3D code can be found in Biedron and Thomas (2009b, 2009c).

##### 3.2.2. FUN3D grids

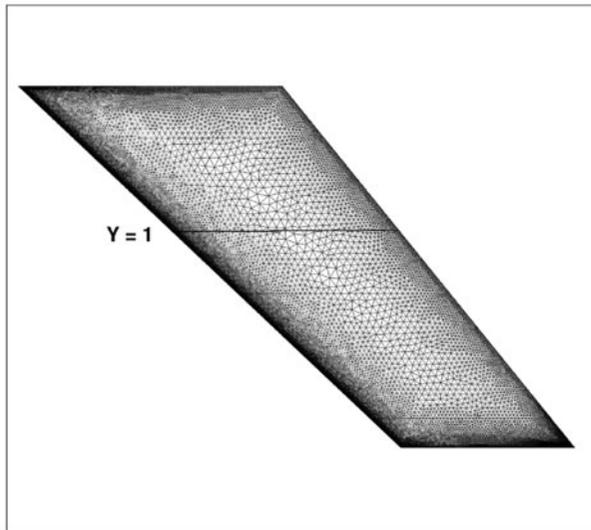
Unstructured tetrahedral grids used in this study were generated using VGRID (Pirzadeh 2008) with input prepared using GridTool (Samareh 1993). For the AGARD 445.6 wing grids, the boundary layer consisted of tetrahedral elements. Only two grids were used for the present analyses: an inviscid (Euler) grid consisting of two million nodes and a viscous (NS) grid consisting of four million nodes, also referred to as the baseline grid. Figure 8 shows the planform, surface grids, and the surface  $C_p$  at  $M = 1.141$  for the inviscid and viscous grids used in this analysis. A relatively strong compression near the trailing edge is present in the inviscid solution. Although the flow is entirely supersonic, it is not surprising to see a strong compression as a result of the inviscid (Euler) analysis.

Note that the surface grid for both the inviscid and viscous grids is the same with the difference in grid dimensions accounting for a denser grid normal to the surface for the viscous grid (not visible in this figure). The surface  $C_p$  at  $M = 1.141$  for the viscous solution indicates that the inclusion of viscosity has diminished the strong compression seen for the inviscid solution.

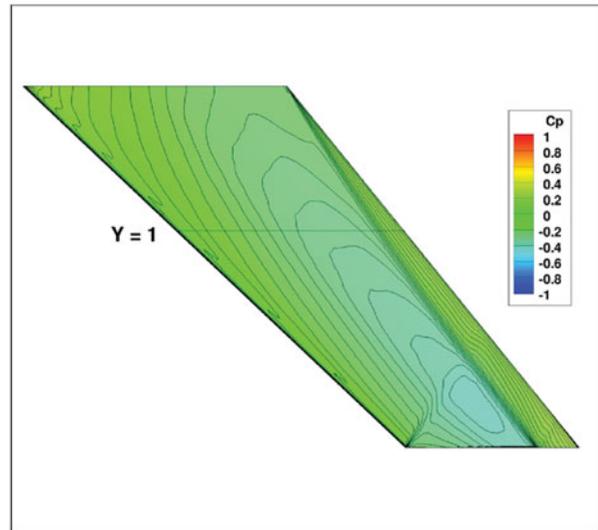
##### 3.2.3. FUN3D analysis procedure

Solutions to the Reynolds-averaged Navier–Stokes (RANS) equations were computed using the FUN3D flow solver. The FUN3D solutions presented in this paper were obtained with an augmented van Leer limiter (Vatsa and White 2009), low-diffusion flux-splitting scheme (LDFSS) (Edwards 1995) for inviscid fluxes, and the SA turbulence model (Spalart and Allmaras 1994). For the asymptotically steady cases under consideration, time integration was accomplished by an Euler implicit backwards difference scheme, with local time stepping to accelerate convergence. Most of the steady-state cases in this study were run for 5000 iterations to achieve convergence of forces and moments to within  $\pm 0.5\%$  of the average of their last 1000 iterations.

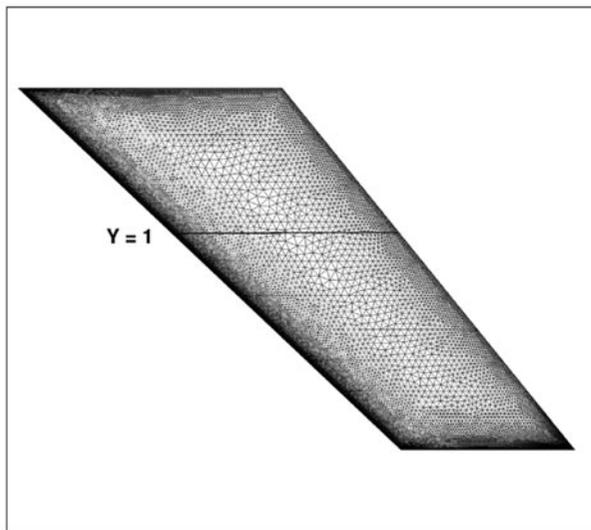
In order to perform static and dynamic aeroelastic solutions, interpolation of the structural mode shapes onto the CFD surface grid is required. This interpolation is done as a preprocessing step (Samareh 2007). The final surface deformation at each time step is a linear superposition of all the modal deflections.



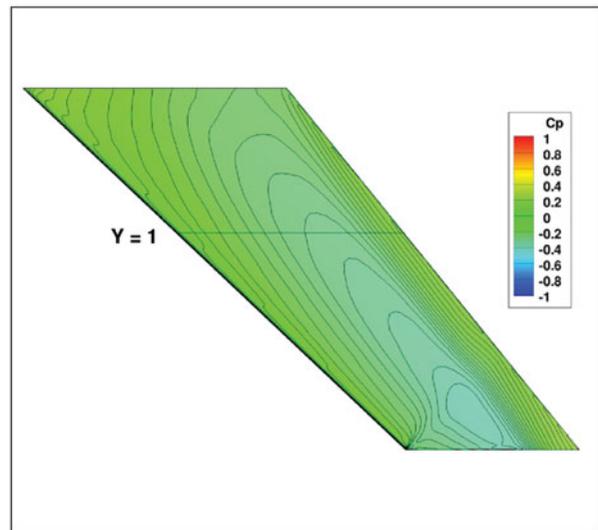
(a) Inviscid FUN3D surface grid for the 445.6 wing.



(b) Pressure coefficient contours on the surface for the inviscid FUN3D solution.



(c) Viscous surface grid for the 445.6 wing.



(d) Pressure coefficient contours on the surface for the viscous FUN3D solution.

Figure 8. Inviscid and viscous grids and pressure distributions of the AGARD 445.6 wing at  $M = 1.141$ .

The standard procedure for performing a FUN3D full aeroelastic solution was performed as follows. First, the steady CFD solution was obtained on the rigid vehicle. Next, a static aeroelastic solution was obtained by continuing the CFD analysis in a time accurate mode, allowing the structure to deform but with a high value of structural damping (0.99) so the structure could find its equilibrium position with respect to the mean flow before the dynamic response was started. However, for the AGARD 445.6 wing, there was no need to compute static aeroelastic results due to the fact that the airfoil is symmetric and the analyses are

performed with the wing at  $0^\circ$  angle of attack. Finally, for the dynamic response, the damping was set to an assumed value (0.00), and the structure was perturbed in generalized velocity for each of the four modes included in the structural model. The flow was then solved in the time accurate mode.

### 3.3. FUN3D reduced-order models

In this section, the FUN3D reduced-order model (ROM) development process is briefly reviewed with the details being deferred to the references.

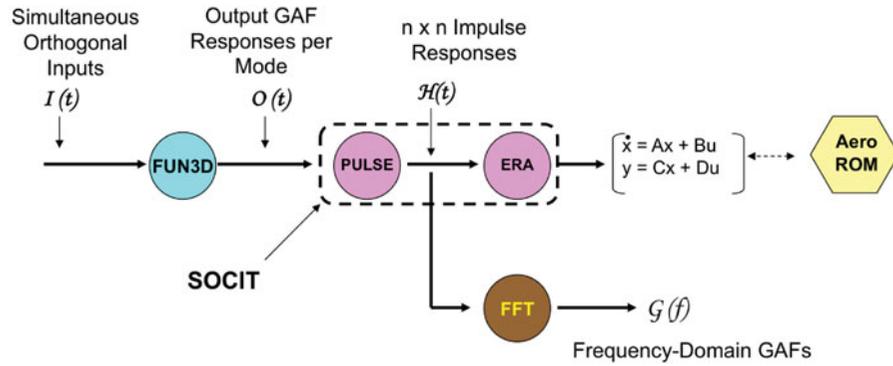


Figure 9. Improved process for generation of an unsteady aerodynamic ROM (Steps 1–4).

An outline of the FUN3D ROM development process is as follows:

- (1) Generate the number of functions from a selected family of orthogonal functions (Silva 2008) that corresponds to the number of structural modes.
- (2) Apply the generated input functions simultaneously via one CFD execution resulting in GAF responses due to these inputs; these responses are computed directly from the restart of a steady rigid CFD solution (not about a particular dynamic pressure).
- (3) Using the simultaneous input/output responses, identify the individual impulse responses using the PULSE algorithm within SOCIT (Juang 1993).
- (4) Transform the individual impulse responses generated in Step 3 into an unsteady aerodynamic state-space system using the ERA (Juang and Pappa 1985) (within SOCIT).
- (5) Evaluate/validate the state-space models generated in Step 4 via comparison with CFD results (i.e., ROM results vs. full CFD solution results).

A schematic of Steps 1–4 of the process outlined above is presented as Figure 9. Using modal information (generalized mass, frequencies, and dampings), a state-space model of the structure is generated. This state-space model of the structure is referred to as the structural state-space ROM (Figure 10). Once an unsteady aerodynamic ROM and a structural state-space ROM have been generated, they

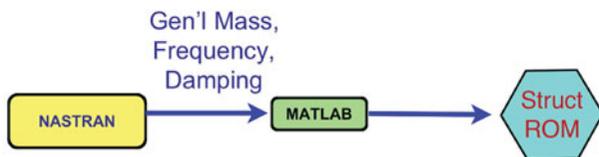


Figure 10. Process for generation of a structural state-space ROM.

are combined to form an aeroelastic simulation ROM (see Figure 11). Then, root locus plots are extracted from the aeroelastic simulation ROM.

## 4. Results

### 4.1. Linear CAP-TSD results

In this section, linear CAP-TSD results are presented at three Mach numbers:  $M = 0.90, 0.96,$  and  $1.141$ . Results are presented in the form of aeroelastic transients for different dynamic pressures and a root locus plot per Mach number. The ROM method has not been incorporated into the CAP-TSD code at this point in time. Instead, the aeroelastic transients are analysed for damping and frequency content using a new MATLAB-based version of a procedure

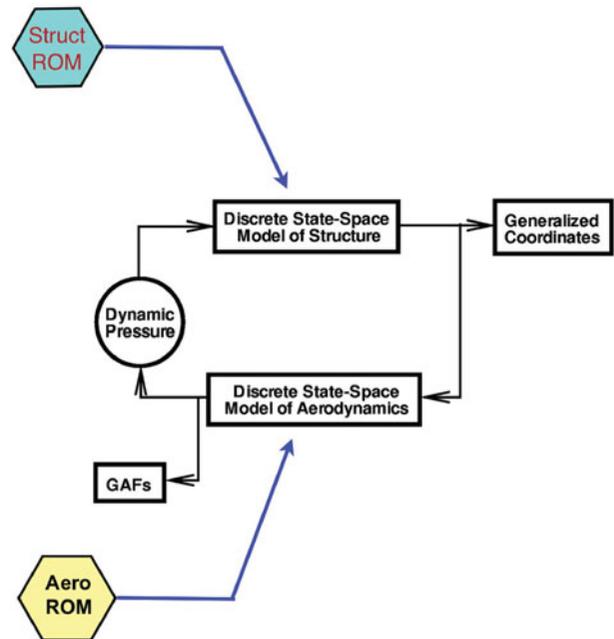


Figure 11. Process for generation of an aeroelastic simulation ROM consisting of an unsteady aerodynamic ROM and a structural state-space ROM.

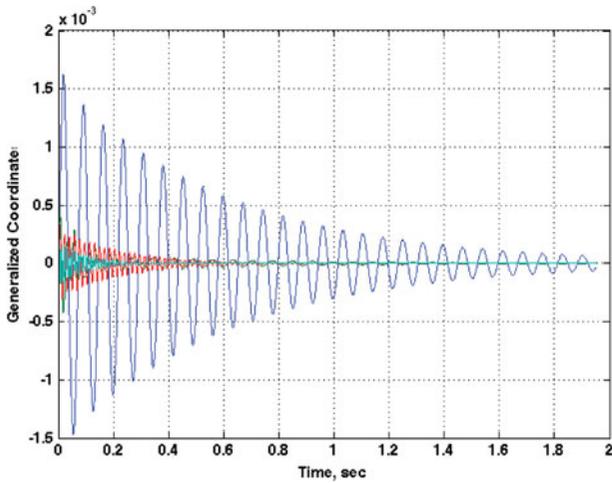


Figure 12. Linear CAP-TSD generalized coordinates at  $M = 0.90$ ,  $Q = 50$  psf

originally developed by Bennett and Desmarais (1975). This algorithm applies MATLAB's curve-fitting routines along with MATLAB's optimization routines in order to find the best curve fit for a given aeroelastic transient consisting of amplitude, damping, and frequency for up to three combined sinusoidal functions.

For the sake of brevity, a small number of aeroelastic transients are analysed per Mach number using this new algorithm. For example, four dynamic pressures were analysed for damping and frequency content for the  $M = 0.90$  results, three dynamic pressures were analysed for damping and frequency content for the  $M = 0.96$  results, and five dy-

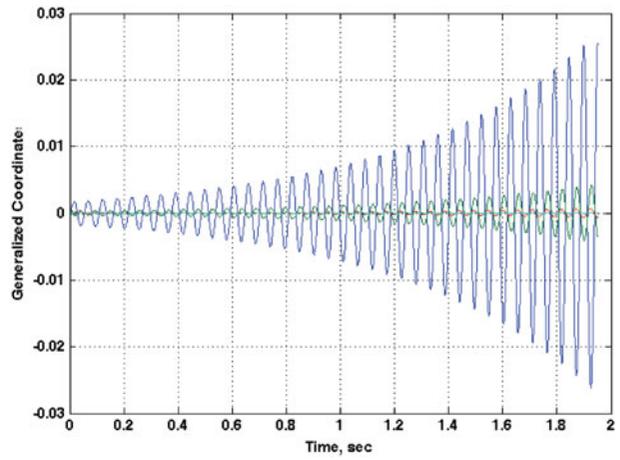


Figure 13. Linear CAP-TSD generalized coordinates at  $M = 0.90$ ,  $Q = 100$  psf

amic pressures were analysed for damping and frequency content for the  $M = 1.141$  results. Additional damping and frequency estimates could be generated at additional dynamic pressures per Mach number in order to generate denser root locus plots. However, as long as the primary flutter mechanism was evident, a need for additional damping and frequency estimates was deemed unnecessary.

#### 4.1.1. Mach number 0.9

Figure 12 presents the linear CAP-TSD generalized coordinates for all four modes at  $M = 0.90$  and a dynamic pressure of 50 psf. As is clear from the figure, this condition is stable.

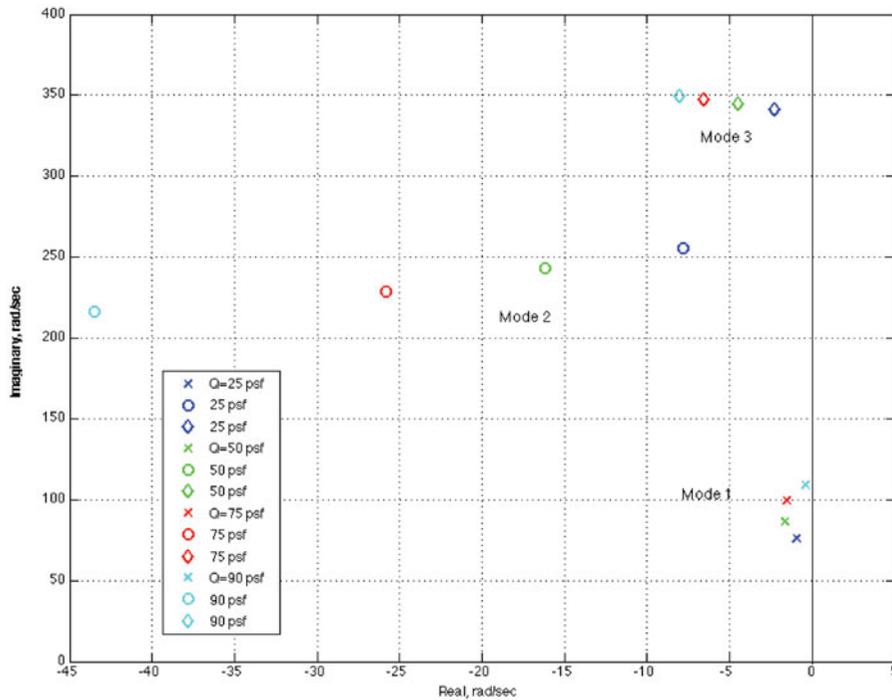


Figure 14. Linear CAP-TSD aeroelastic root locus for  $M = 0.90$ .

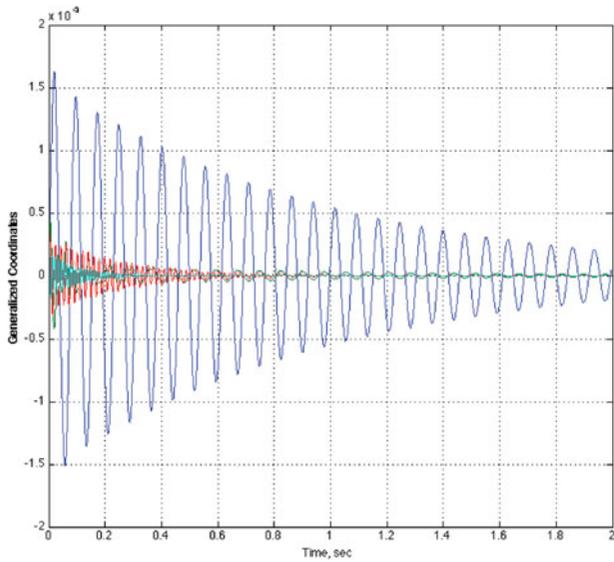


Figure 15. Linear CAP-TSD generalized coordinates at  $M = 0.96$ ,  $Q = 50$  psf

Figure 13 presents the linear CAP-TSD generalized coordinates for all four modes at  $M = 0.90$  and a dynamic pressure of 100 psf. At this condition, an unstable flutter condition is evident. Figure 14 presents the aeroelastic root locus as a function of dynamic pressure for  $M = 0.90$ . This root locus plot indicates that flutter occurs just above 90 psf and that the flutter mechanism is dominated by the first mode participation.

4.1.2. Mach number 0.96

Figure 15 presents the linear CAP-TSD generalized coordinates for all four modes at  $M = 0.96$  and a dynamic pressure of 50 psf. As is clear from the figure, this condition is stable. Figure 16 presents the linear CAP-TSD generalized coordinates for all four modes at  $M = 0.96$  and a dynamic pressure of 90 psf. At this condition, an unstable flutter condition is evident. Figure 17 presents the

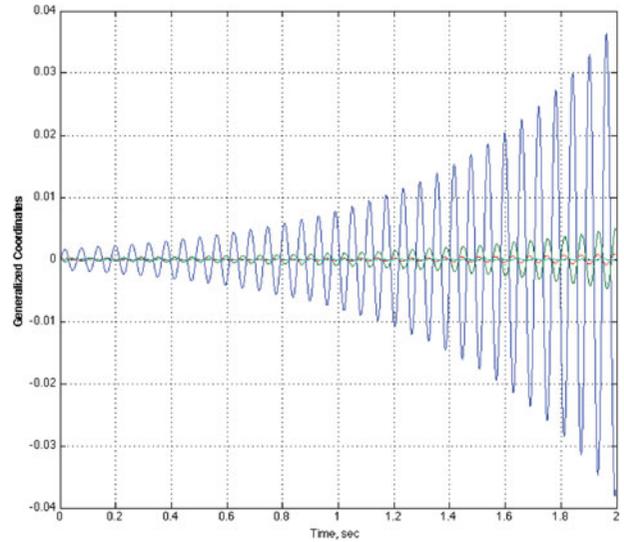


Figure 16. Linear CAP-TSD generalized coordinates at  $M = 0.96$ ,  $Q = 90$  psf

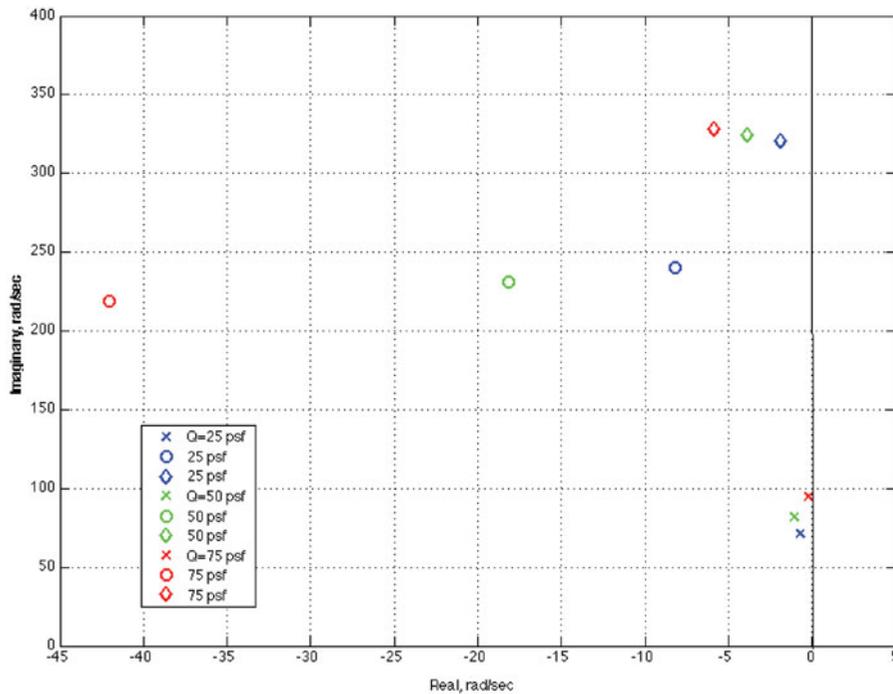


Figure 17. Linear CAP-TSD aeroelastic root locus for  $M = 0.96$

aeroelastic root locus as a function of dynamic pressure for  $M = 0.96$ . This root locus plot indicates that flutter occurs just above 75 psf and that the flutter mechanism is, again, dominated by the first mode participation. For this case, the MATLAB-based damped sine curve-fitting function had some difficulty in providing good estimates of damping for an unstable transient.

4.1.3. Mach number 1.141

Figure 18 presents the linear CAP-TSD generalized coordinates for all four modes at  $M = 1.141$  and a dynamic

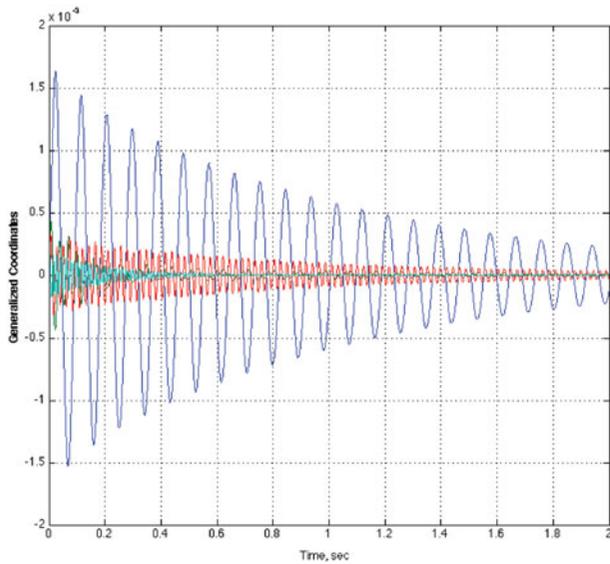


Figure 18. Linear CAP-TSD generalized coordinates at  $M = 1.141$ ,  $Q = 50$  psf

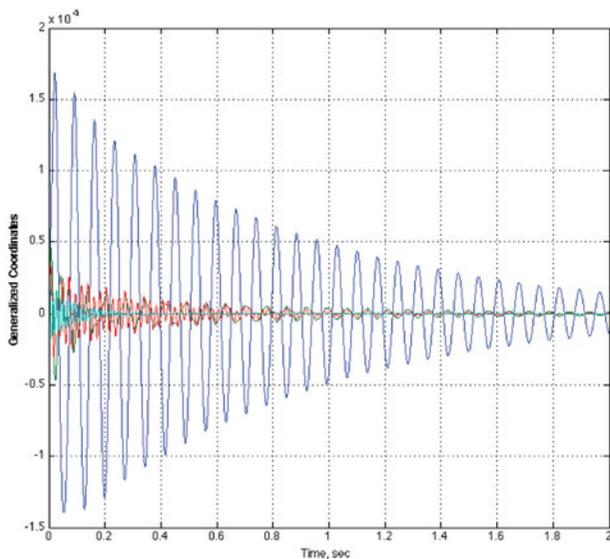


Figure 19. Linear CAP-TSD generalized coordinates at  $M = 1.141$ ,  $Q = 100$  psf

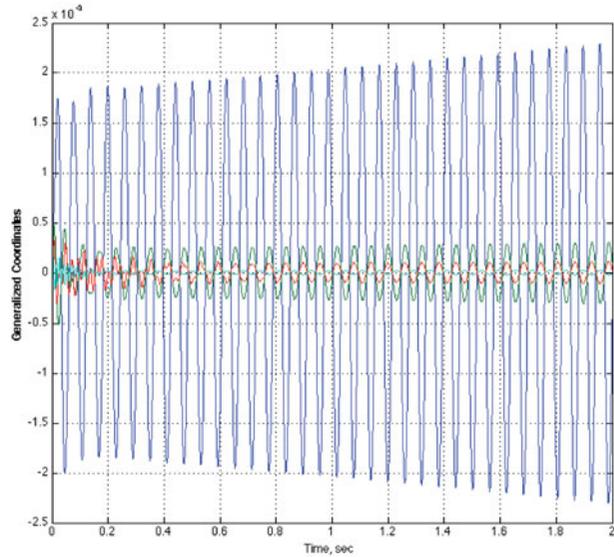


Figure 20. Linear CAP-TSD generalized coordinates at  $M = 1.141$ ,  $Q = 140$  psf

pressure of 50 psf. As is clear from the figure, this condition is stable. Figure 19 presents the linear CAP-TSD generalized coordinates for all four modes at  $M = 1.141$  and a dynamic pressure of 100 psf. This condition is stable as well. Figure 20 presents the linear CAP-TSD generalized coordinates for all four modes at  $M = 1.141$  and a dynamic pressure of 140 psf. Clearly, at this condition, an unstable flutter condition is evident. Figure 21 presents the aeroelastic root locus as a function of dynamic pressure for  $M = 1.141$ . This root locus plot indicates that the flutter occurs at about 140 psf. Once again, the flutter mechanism is dominated by the first mode participation.

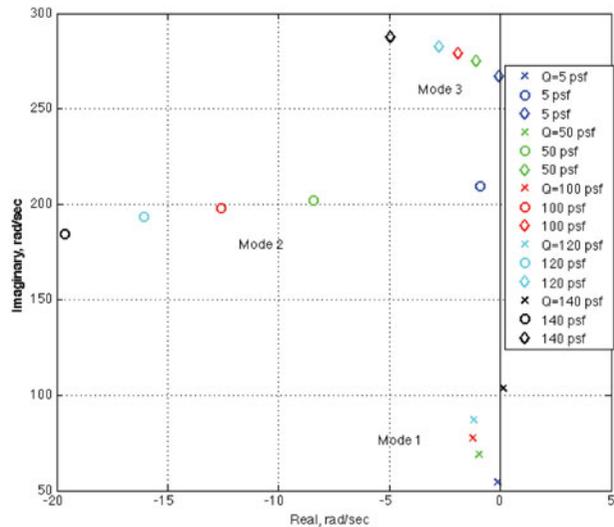


Figure 21. Linear CAP-TSD aeroelastic root locus for  $M = 1.141$

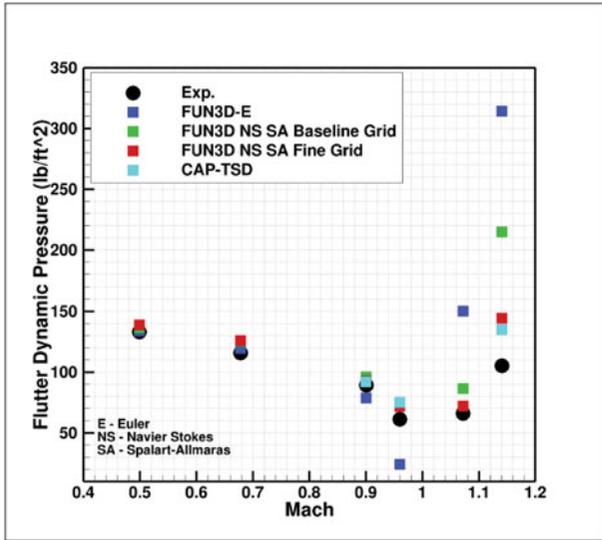


Figure 22. Flutter boundaries as a function of dynamic pressure (psf) including linear CAP-TSD results.

Flutter dynamic pressures previously shown in Figure 6, but now modified to include the linear CAP-TSD results, are presented in Figure 22. As mentioned before and as presented by Cunningham, Batina, and Bennett (1988) and Bennett, Batina, and Cunningham (1989), the linear CAP-TSD code shows excellent agreement with experiment for the subsonic Mach numbers. For the supersonic Mach number, the linear CAP-TSD result is above (not conservative) the experimental flutter dynamic pressure, as are all the FUN3D results, but it is closer to the experimental value than any of the FUN3D results presented.

Flutter frequencies corresponding to the flutter dynamic pressures presented in Figure 22 are presented in Figure 23.

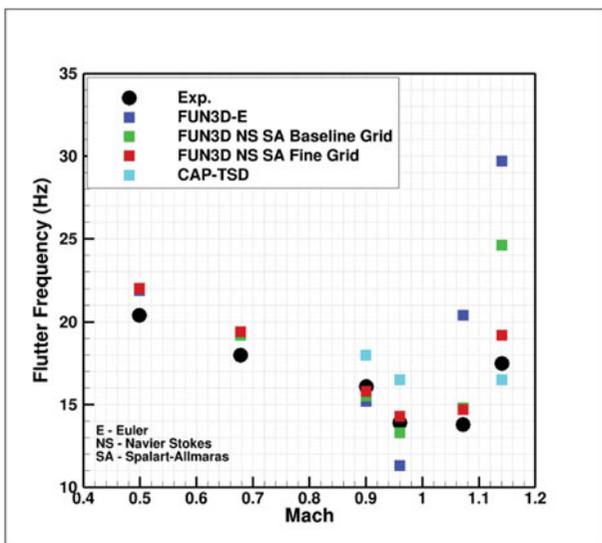


Figure 23. Flutter frequencies (Hz) including linear CAP-TSD results.

For the subsonic Mach numbers, the linear CAP-TSD flutter frequencies are higher than the experimental and the FUN3D flutter frequencies. However, for the supersonic condition, the linear CAP-TSD flutter frequency is closer to the experimental value than the FUN3D results.

#### 4.2. FUN3D full and ROM results

In this section, aeroelastic transients and aeroelastic root locus plots are presented for the FUN3D full and ROM results for both inviscid and viscous solutions. The FUN3D full results consist of aeroelastic transients at various dynamic pressures for two Mach numbers:  $M = 0.96$  and  $M = 1.141$ . The FUN3D ROM results will consist of aeroelastic root locus plots for the same Mach numbers. The root locus plots generated using the ROM procedure described above are used to identify aeroelastic behaviour and flutter mechanisms. The aeroelastic transients generated using the FUN3D full solutions are used to validate the FUN3D ROM results at specific dynamic pressures.

##### 4.2.1. Inviscid results

In this section, inviscid FUN3D results are presented for both full and ROM solutions. Presented in Figure 24 is the aeroelastic root locus plot for  $M = 0.96$  using the FUN3D ROM method. In contrast to the root locus plots presented for the linear CAP-TSD solutions, these root locus plots contain root values at 20 dynamic pressures from 0 to 114 psf. The reason for this increased resolution in dynamic pressure values is the ROM procedure itself. In this case, there is no need to analyse an aeroelastic transient at each and every dynamic pressure as was the case for the linear CAP-TSD results. Because the ROM procedure generates a combined aeroelastic state-space model that consists of a state-space model of the structure and a state-space model of the unsteady aerodynamics (from FUN3D), root locus plots can be generated for any number and any increment of dynamic pressure in a matter of seconds. The flutter mechanism for the FUN3D inviscid solution at this Mach number is clearly dominated by the first mode with some coupling with the second mode noticeable. The third and fourth modes are stable.

In order to better visualize the root migrations for the first mode, a zoomed-in version of the root locus plot is presented as Figure 25. The increment in dynamic pressure for this root locus plot is 6 psf starting with 0 psf yielding a flutter dynamic pressure of approximately 30 psf. This result is very close to and consistent with the FUN3D full solution flutter dynamic pressure presented in Figure 6. However, the inviscid result at this Mach number does not compare well with the experiment. This is not surprising as inviscid solutions tend to have stronger shocks that are farther aft, and therefore, induce a stronger and earlier onset of flutter. When viscosity is introduced into the solution, the

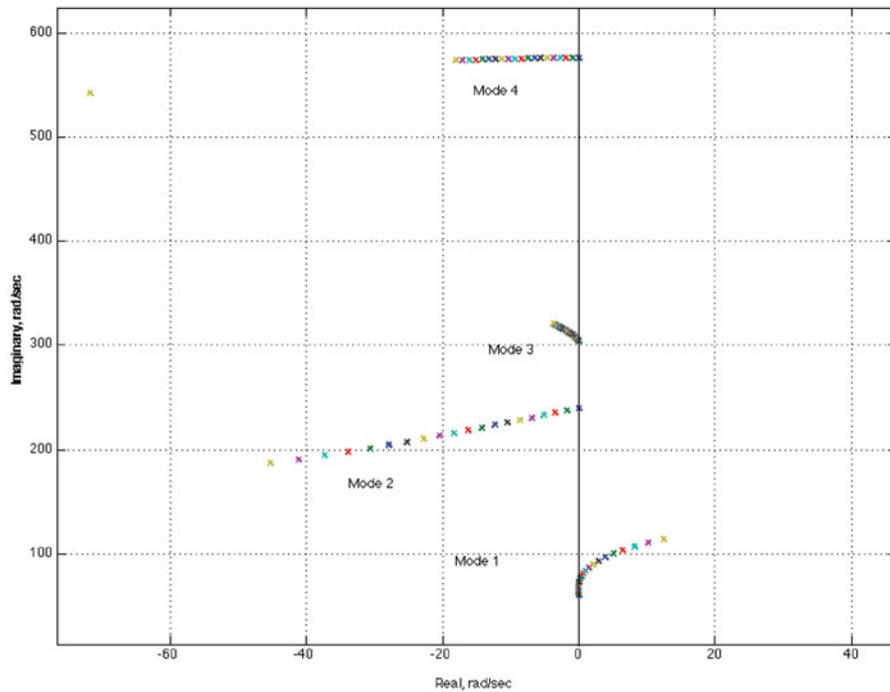


Figure 24. FUN3D ROM aeroelastic root locus plot for  $M = 0.96$ , inviscid solution.

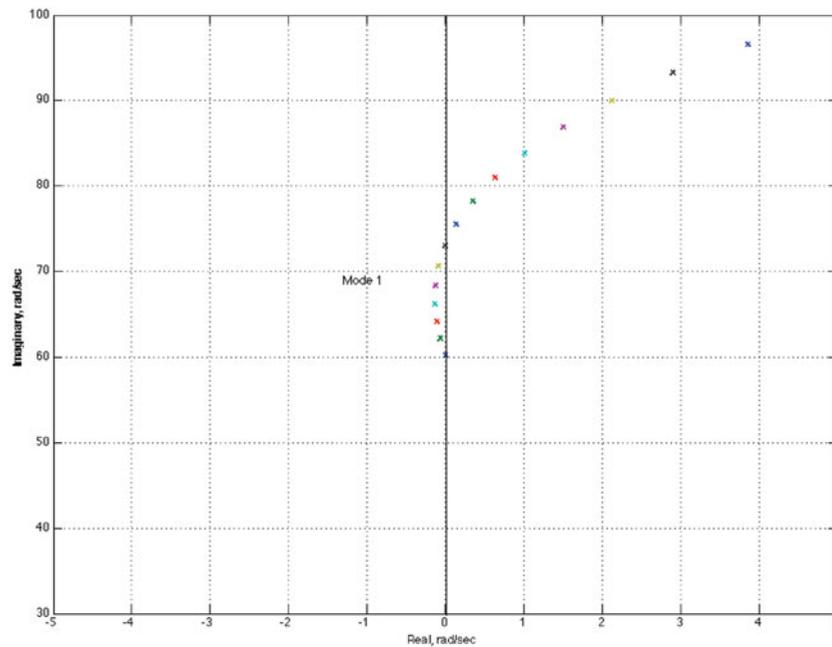


Figure 25. Zoomed-in version of FUN3D ROM aeroelastic root locus plot for  $M = 0.96$ , inviscid solution.

shock strength is reduced and the shock position is moved forward resulting in the onset of flutter at a higher dynamic pressure. This effect is discussed in the next section of this paper.

Solutions are now presented for the supersonic Mach number of 1.141. As presented in Figure 6, there is a wide variation of flutter dynamic pressures and flutter frequencies at this condition. The discrepancy between many of

the solutions and the experiment as well as the discrepancy amongst the various solution methods has long been a source of speculation. Although the authors do not present a conclusive answer to the source of these discrepancies, it is hoped that the results presented will spur additional discussion and research.

Figure 26 presents the aeroelastic root locus plot for  $M = 1.141$  generated using the FUN3D ROM method.

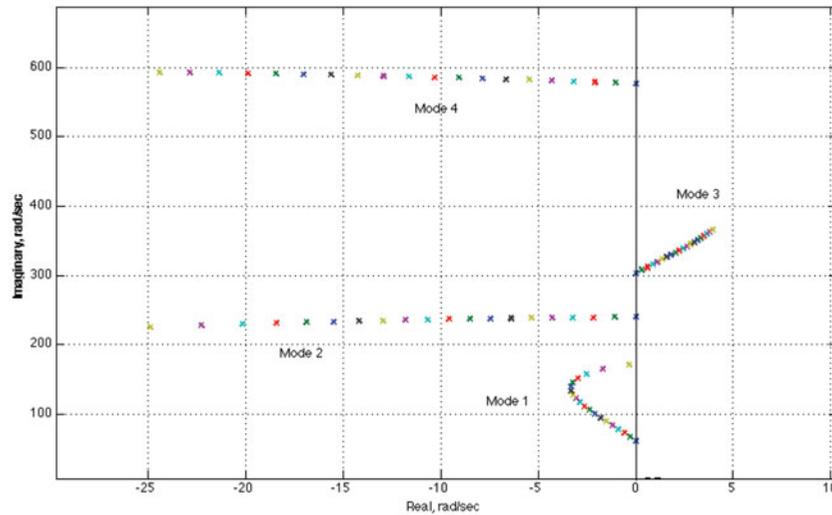


Figure 26. FUN3D ROM aeroelastic root locus plot for  $M = 1.141$ , inviscid solution.

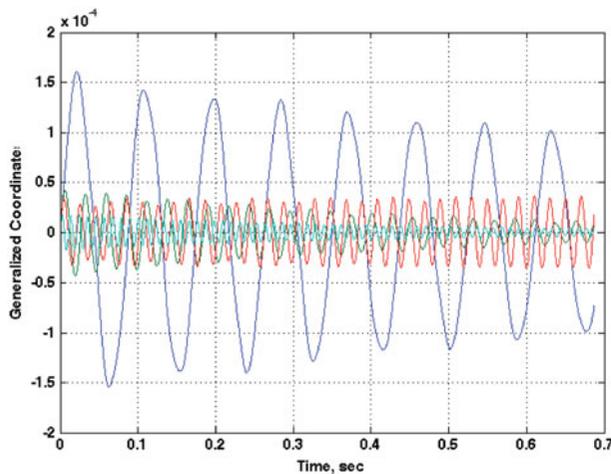


Figure 27. FUN3D full solution generalized coordinates at  $M = 1.141$ ,  $Q=30$  psf, inviscid solution.

There are two obvious flutter mechanisms: one flutter mechanism is an instability involving the first mode while the other flutter mechanism involves an instability of the third mode. The third mode is always unstable while the first mode instability occurs at a dynamic pressure of about 300 psf. The third mode instability was a surprise in that it is not mentioned by other researchers. Therefore, in order to validate the accuracy of this aeroelastic root locus plot, the generalized coordinates from a FUN3D full solution are inspected.

Presented in Figure 27 are the aeroelastic transients for the four modes at  $M = 1.141$  and a dynamic pressure of 30 psf. The first mode, with the largest amplitude, is clearly stable. However, discerning the stability for the other three modes with smaller and similar amplitudes is not as obvious as it is for the first mode. If only the third mode is visualized, as in Figure 28, the unstable nature of this mode becomes clearer.

This third mode instability, not mentioned in any other publications on the flutter boundary of the AGARD 445.6 wing, raises an important question. Is this third mode instability present in all inviscid (Euler) solutions of the AGARD 445.6 wing presented in the literature? The answer to this question requires consultation with researchers that have performed inviscid CFD aeroelastic analyses for this wing at this condition. However, as mentioned previously, it appears that the focus of all previous inviscid analyses at supersonic conditions was the first mode instability. If that is the case, it is easy to see how the third mode instability might have been ignored. In addition, for analyses performed in the early days of computational aeroelasticity, Figure 27 would have consisted of fewer time steps (due to computational cost at the time), thereby making it difficult to visually notice the third mode instability. It should be stated that the authors have confirmed the existence of this third mode instability in previous solutions obtained using the CFL3D code.

#### 4.2.2. Viscous results

In this section, viscous FUN3D full and ROM solutions are presented at  $M = 1.141$  since significant discrepancies, as just discussed, exist at this condition. The results for FUN3D full and ROM solutions at subsonic Mach numbers agree well with each other and with experiment, and are not presented in this paper.

Presented in Figure 29 is the root locus plot generated using the FUN3D ROM viscous solution at  $M = 1.141$  in dynamic pressure increments of 6–114 psf. It appears that the third mode instability exhibited by the inviscid solution has been stabilized by the inclusion of viscous effects.

Presented in Figure 30 is a zoomed-in version of Figure 29 with a focus on the root migration of the first mode.

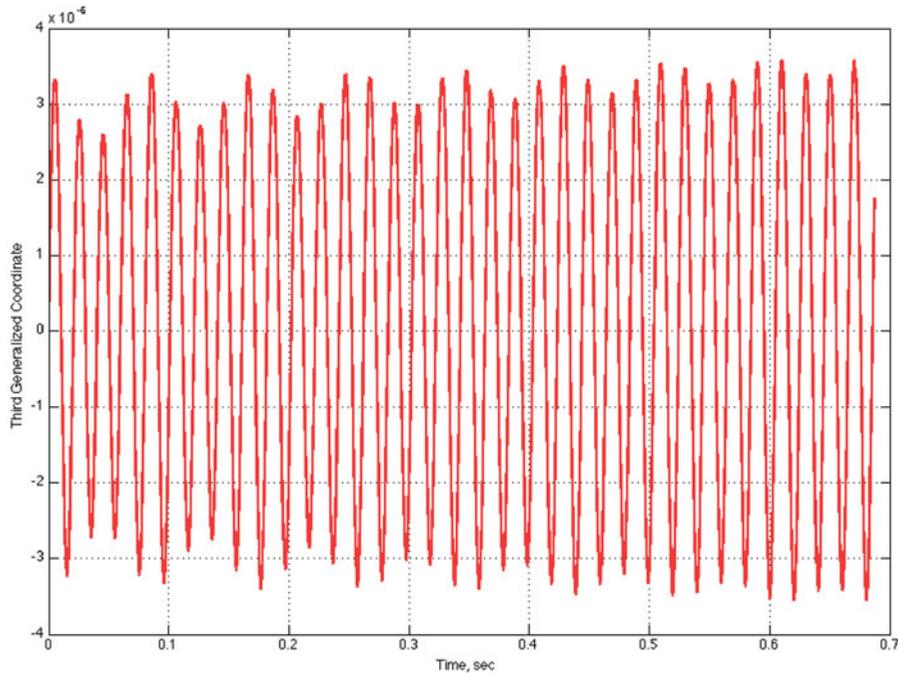


Figure 28. FUN3D full solution third generalized coordinate at  $M = 1.141$ ,  $Q = 30$  psf, inviscid solution.

As can be seen, the first mode is stable throughout these dynamic pressures. However, a zoomed-in version of Figure 29 with a focus on the root migration of the third mode, presented in Figure 31, indicates that initially, the third mode exhibits a slight instability before becoming more stable.

The four generalized coordinates from a FUN3D full viscous solution, at  $Q = 6$  psf, are presented as Figure 32. At this low dynamic pressure, all four generalized coordinates are lowly damped. Figures 33 and 34 present the first and third generalized coordinates from Figure 32, respectively. Visual analysis of both of these generalized coordinates

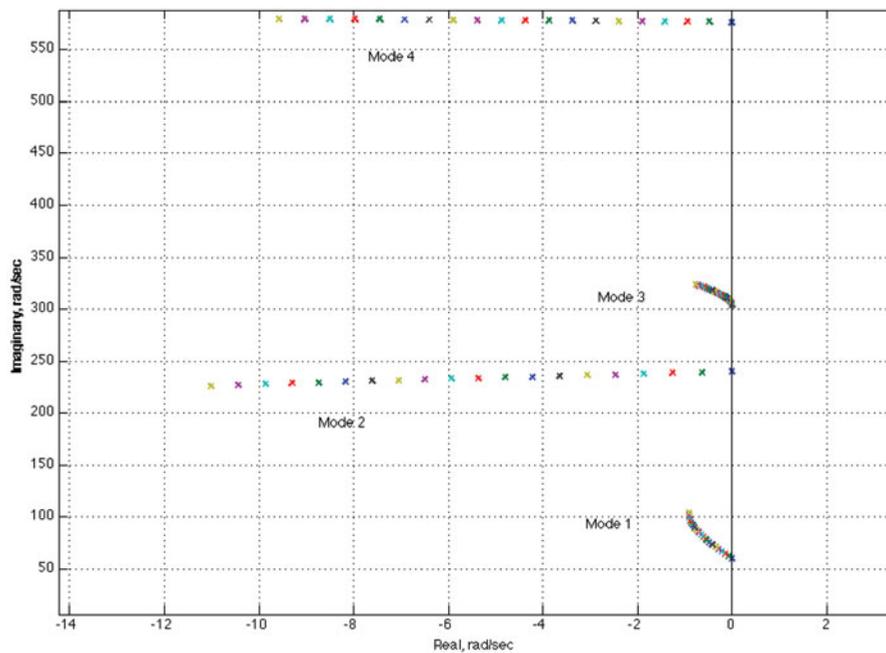


Figure 29. Viscous FUN3D ROM root locus plot at  $M = 1.141$ .

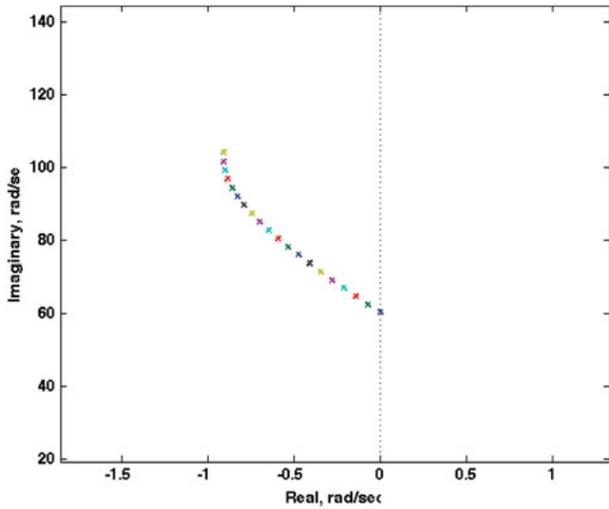


Figure 30. Viscous FUN3D ROM root locus plot at  $M = 1.141$ , zoomed-in to the first mode.

indicates that these generalized coordinates appear to be stable, although lowly damped. However, it is important to state a fundamental and important difference between a root locus plot and the visual, or otherwise post-processed, analysis of generalized coordinates over a short period of time. A root locus plot, by definition, exhibits the roots of a system as time approaches infinity or as the system reaches steady state. In contrast, the analysis of the initial transient response of a generalized coordinate over a short period of time can be deceiving as the response can change if the response was viewed (or analysed) over a longer period of time. Therefore, based on the root locus plot at this condition, it appears that the third mode is unstable although that is not apparent in Figure 34.

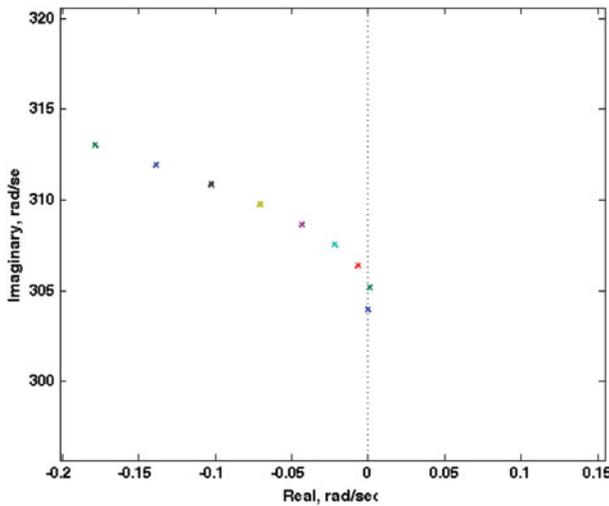


Figure 31. Viscous FUN3D ROM root locus plot at  $M = 1.141$ , zoomed-in to the third mode.

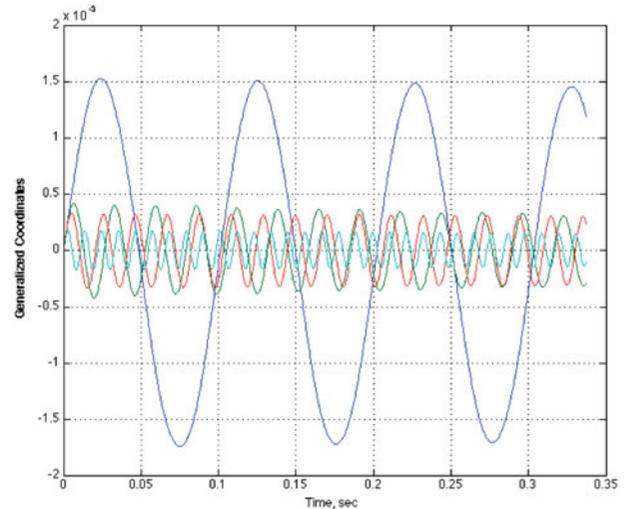


Figure 32. Generalized coordinates from viscous FUN3D full solution at  $M = 1.141$  and  $Q = 6$  psf.

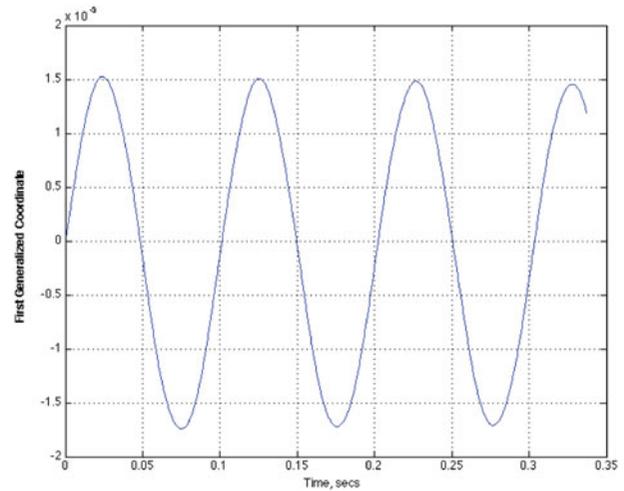


Figure 33. First generalized coordinate from viscous FUN3D full solution at  $M = 1.141$  and  $Q = 6$  psf.

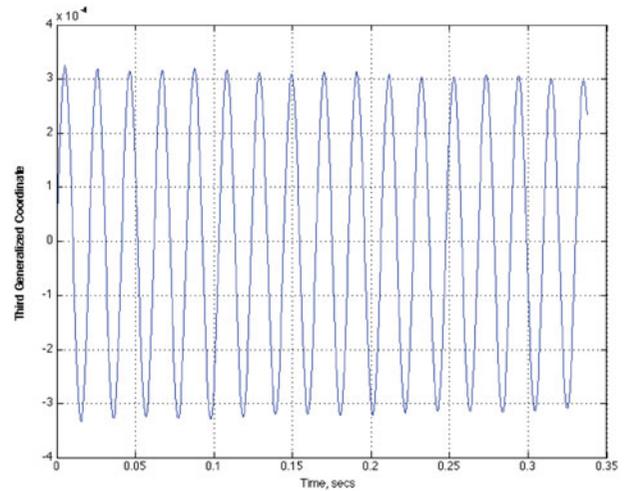


Figure 34. Third generalized coordinate from viscous FUN3D full solution at  $M = 1.141$  and  $Q = 6$  psf.

## 5. Concluding remarks

A comparison of linear, inviscid, and viscous aeroelastic solutions for the AGARD 445.6 wing was presented. The linear solutions were generated using the linear equations available within the CAP-TSD code. The inviscid and viscous solutions were generated using the FUN3D code. The results presented consisted of two types of solutions: full CFD solutions whereby the CFD aeroelastic solution was computed in the standard time iterative approach and ROM solutions whereby unsteady aerodynamic and aeroelastic ROMs were generated using FUN3D and used to generate aeroelastic root locus plots as a function of dynamic pressure. An important conclusion from this research is that visualization of the aeroelastic root locus plots provided by the ROM approach enables a direct and more comprehensive interpretation of the aeroelastic behaviour. Case in point, for the inviscid (Euler) solutions, the third mode is always unstable while the flutter instability associated with the first mode does not occur until about 300 psf. If this third mode instability is present in all inviscid solutions published by other researchers is not known at this point in time. However, the importance of being able to view a root locus plot in addition to time domain responses of the generalized coordinates was well established with this particular result. Another related conclusion to be made is that the standard approach of viewing (and analysing) a short time history of the initial transient response of generalized coordinates may not be sufficiently accurate since it is possible that the steady-state response has not been reached over this short time interval. Here again, the use of the root locus plots generated using the ROM approach enable the complete observation of the aeroelastic response since, by definition, the root locus plots represent the dynamics of the system at the final steady state of the system.

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